# Design Log

## Requirements

### Orbiter

Pointing direction:

* For communications the orbiter must be nadir pointing, except during probe deployment. **Driving requirement**
* Earth-pointing comms will have own gimbal mechanism for increased accuracy and to allow independent pointing direction
* The EPS must therefore be equipped with a gimbal mechanism to allow pointing towards the earth throughout the orbit.

Pointing Accuracy:

* EPS: max 23 degree incidence angle. Will be outfitted with gimbal mechanism, so not a driving req for AOCS
* Telecomms: For high-gain antenna 0.1-0.5 degree accuracy minimum. HGA will utilize separate gimbal for higher accuracy. **Driving requirement**

Maneuver Requirements:

* No need for high-rate maneuvers. Nominal rates will suffice (0.05 deg/s – 0.5 deg/s)
* Range – all angles must be accessible, as the probes must be deployed from various attachment points
* Jitter – unknown at this stage from other sub-systems
* Settling Time – unknown at this stage from other sub-systems

### Probe

Pointing direction:

* Comms will utilize and omni-directional antenna. No ACS requirements
* EPS will rely on internal batteries
* Propulsion – retrograde during de-orbit burn
* Thermal – prograde during atmospheric entry
* EDL – trajectory guidance if using a lifting trajectory
* De-tumble after deployment

Pointing accuracy:

* Exact pointing accuracies unknown at this design stage
* Dependent
* Large deadbands to save fuel [MSL entry]

Maneuver requirements:

* 180 degree flip maneuver during after de-orbit burn, prior to atmospheric re-entry
* Maneuver rate TBD
* Lift/bank control (likely if lifting trajectory is chosen)

## ACS Type

### Orbiter

* Primarily dependent on TT&C pointing requirements and accuracy

Orbit control type will be (SMAD Table 11-4):

* Three-axis, zero-momentum (3 wheels + thrusters)
* High pointing accuracy
* Combination of thrusters and reaction wheels
* Thrusters used for slewing and momentum dumping
* RW for high accuracy pointing
* Control Moment Gyros (CMG) likely not needed, TBD in hardware selection phase

### Probe

* If lifting trajectory is used, spin-stabilized will not be applicable.
* Spin stabilization also unlikely due to the need to execute a 180deg flip maneuver between de-orbit burn and atmospheric re-entry

Control type will be (referring to SMAD table 11-4):

* Three-axis, zero momentum (thruster only)
* Pointing accuracy 0.1 – 5 deg
* High rates possible

## Disturbance Torques

### Orbiter (TODO: get MRO data)

* Solar
  + Assume reflectance factor of q = 0.6. Based on SMAD
  + Incidence angle assumed to be worst case, i = 0
  + Cp and Cg are co-incident at this phase, producing no torques
  + **Negligible at this phase**
* Gravity Gradient
  + **Negligible at this phase**
  + S/C will be nadir pointing, and will thus gravity gradient will work as an advantage (IF slender shape points aligned along z-axis
  + This design phase, s/c is a symmetrical cube
* Aerodynamic Drag
  + Placement of appendages is unknown, cannot determine resultant Cp
  + Current design: CG and CP both located in center, so resultant torque is zero
  + **Negligible at this phase**
* Magnetic
  + At Earth GEO, negligible. If magnetic field of mars at the design orbit is less than this, then this disturbance can also be neglected [book:Spacecraft-FCS-design, p1]
  + Earth orbit: mag only of significance in orbits below 1500km. (32,000 nT)
  + Mars magnetic field is max 1500 nT [<https://www.planetary.org/blogs/emily-lakdawalla/2008/1710.html>, retrieved 5/6/2020]
  + **Negligible at this phase**

Because the geometry of the spacecraft is unknown at this phase, it is not possible to calculate disturbance torques. It will be assumed that the slewing torque to execute maneuvers such as probe deployment will be significantly greater than the disturbances, and the reaction wheels will be sized using these torque requirements.

Depending on the torque required, which is determined by the MOI of the orbiter, it may not be feasible to utilize only reaction wheel. As such, thrusters will need to additionally be utilized to exert enough torque to accelerate the spacecraft sufficiently to execute the maneuver in time. Knowing how quickly the orbiter must be able to slew (such as for deployment of probes) will determine thruster sizing.

For this phase it will be assumed that reaction wheels will be used to maintain pointing of the spacecraft, which is needed in any case for the Nadir pointing of LGA antennas towards Mars.

### Probe

* Aerodynamic Drag
* Other torques negligible

The probe will need to be inherently stable throughout the entry and descent phase. The addition of active attitude control is dependent on whether the trajectory will be ballistic or lifting. In the case of lifting trajectory, thrusters will be utilized to keep oscillations within the acceptable deadbands.

The sizing of such thrusters will only be done once it is known which trajectory type is used. Then, the required thrust can be computed based on max AoA excursion, and rough MOI estimates of the probe. Again, the geometry is unknown at this phase, and therefore the aerodynamic properties cannot be estimated.

## Hardware Selection

### Orbiter

* Three-axis control:
  + Thrusters
  + Reaction wheels
  + Control Moment Gyros
  + Combination…

### Probe

* Likely thruster only (see 2007\_Brugarolas on MSL attitude control) wrt high attitude deadband vs low rate deadbands

## Control Law

### Orbiter

### Probe

* See 2013\_SanMartin\_Dev-of-MSL

## Budgets (Basic sizing based on MRO and MSL, **NO CALCULATIONS**)

### Orbiter (selection based on MRO)

* RCS (not MRO) mass TBD
* RWA 2 – 20kg; 10 – 110W (per wheel) [SMAD]
  + MRO 10kg per RW x 4 wheels
  + MRO mass = 1000kg, our orbiter = 1990kg
  + Estimate 20kg per RW 🡪 **80kg** total
* Sensors:
  + Star trackers 2 – 5kg; 5 – 20 W
  + Sun sensors 0.1 – 2kg; 0 – 3 W
  + IMU 1 – 15kg 10 – 200; W
* TOTAL MASS:

### Probe (based on MSL)

* RCS propellant for MSL:
  + Max 15kg for lander of 900kg (so 1.666%)
  + Assuming our probes of 100kg 🡪 2kg propellant to be safe
* Sensors:
  + IMU 1 – 15kg 10 – 200 W

### Other Sub-systems

#### EPS:

Orbiter power range: 25 – 333 W

Probe power range: 10 – 200 W (IMU)

#### Thermal:

Orbiter thermal range:

Probe thermal range:

#### CDH:

Commands from earth:

* s/c attitude model [2007\_You]

Commands from orbiter:

* attitude estimation [2007\_You]

Housekeeping telemetry:

* TBD